

Constants: $M_{\text{sun}} = 2 \times 10^{30} \text{ kg}$ $M_{\text{earth}} = 6 \times 10^{24} \text{ kg}$ $1 \text{ AU} = 1.5 \times 10^{11} \text{ m}$ $R_{\text{sun}} = 700,000 \text{ km}$

- Side 1*

Gravity Problems II

6. A satellite of mass m has a circular orbit of radius R around a planet of mass M . Show that the total energy of the satellite is given by the expression $E = -G \frac{mM}{2R}$. (If you are bored, prove the equation is true even in the case of an elliptical orbit, where R is the semi-major axis. To do this, think about the energy at perihelion and aphelion and also think about conservation of angular momentum.)
7. For some reason, a 5000 kg satellite is put into orbit around the sun with a period of 7 years. What is the total energy of the satellite in this orbit?

Gravity Problems II

8. An asteroid has a period of 2585 days around the sun. If its *maximum* speed in its orbit around the sun is 40,000 m/s, what is its closest distance to the sun (i.e. what is its perihelion?)

Answers:

- 1) 617,000 m/s 2) 29,800 m/s 3) 42,200 m/s
4) 300 km 5) $(2/3)Gm_1m_2/d$
6) *hint 1: total energy = kinetic + potential*
hint 2 (elliptical): E @perihelion = E @aphelion & L is also conserved
7) $-6.1 \times 10^{11} \text{ J}$ 8) $1.44 \times 10^{11} \text{ m}$